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Corrigendum: Coupling plasma physics and chemistry in the PIC model of electric propulsion: application to an air-breathing, low-power Hall thruster

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A Corrigendum on

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In the published article, there was an error in [Table 5](#) as published. In the last column (Anode Mass flow rate) the incorrect value 0.1 mg/s was reported. The correct value is 1.0 mg/s. The corrected [Table 5](#) and its caption appear below.

In the published article, there was an error. The reported mass flow rate of 0.1 mg/s is incorrect.

A correction has been made to **3 Results**. This sentence previously stated:

“For the different propellants analyzed, the mass flow rate is fixed at $\dot{m} = 0.1$ mg/s.”

The corrected sentence appears below:

“For the different propellants analyzed, the mass flow rate is fixed at $\dot{m} = 1.0$ mg/s.”

The authors apologize for these errors and state that these do not change the scientific conclusions of the article in any way. The original article has been updated.

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TABLE 5 Engineering parameters used for the SPT20 simulations.

Engineering parameters	Inner/outer radius (cm)	Channel length (cm)	Discharge voltage (V)	Discharge power (W)	Anode Mass flow rate (mg/s)
Case A: Xe	0.5/1	1	200	56	1.0
Case B: N ₂	0.5/1	1	200	52	1.0
Case C: O ₂	0.5/1	1	200	96	1.0
Case D: N ₂ -O	0.5/1	1	200	84	1.0